

Claims 3, 6, 7, 10-12, 14-18, 20-26 stand cancelled.

Claims 8, 9, 13, 19 and 27-30 stand withdrawn from consideration.

STATUS OF AMENDMENTS

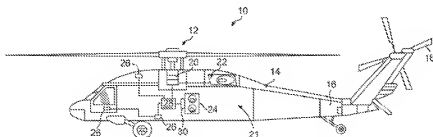
All amendments have been entered.

SUMMARY OF CLAIMED SUBJECT MATTER

The dominant source of vibration in a helicopter is that generated by the main rotor system at the blade passing frequency (rotation rate times the number of rotor blades). Forces and moments are transmitted usually through the transmission via fuselage attachments, to produce vibration in the fuselage. [Page 1, lines 8-11]

One conventional approach to reducing such vibration involves replacing a rigid gearbox mounting strut with a compliant strut and parallel hydraulic actuator. Another conventional approach utilizes force generators consisting of counter-rotating eccentric masses that rotate at the frequency of the primary aircraft vibration and generate a fixed magnitude vibration force. This system, although effective for direct gearbox mounting, requires a parasitic mass of considerable magnitude which may result in an unacceptable weight penalty. [Page 1, lines 12-22]

The present application relates to producing large, controllable, vibratory forces to compensate for sensed noise or vibrations, and more particularly to a force generator which is part of an active vibration control (AVC) system for an aircraft. [Page 1, lines 5-7]



In operation, vibratory forces are produced by the main rotor assembly 12 due, for example, to asymmetric air flow in forward flight. Such vibratory forces arising as the main rotor assembly 12 rotates are, in the absence of any compensating systems, transmitted from the

rotor 12 to the fuselage 14. Operation of the force generator(s) 24 is continuously varied by the processor 28 to cater to changing dynamic characteristics such that vibratory forces caused by the rotor assembly 12 and/or other vibratory sources are reduced or eliminated. [Page 3, line 29 – Page 4, line 4]

Summary of Claim 1

Referring in particular to Figures 2A and 2B, claim 1 recites a processor 28 that controls a power source 30 to drive a crank 38 such that a phase and magnitude of the vibratory inertial force is continuously varied to reduce an externally generated vibratory force sensed by a sensor system 26 [Page 3, line 29 – Page 4, line 5].

GROUND S OF REJECTION TO BE REVIEWED ON APPEAL

A. 35 U.S.C. §103

1. Were claims 1, 2, 4, 5, 31, and 32 properly rejected under 35 U.S.C. §103(a) as being unpatentable over *Kanski*. (2309172) in view of *Maslov* (US 20050184689).

2. Were claims 33-36 properly rejected under 35 U.S.C. §103(a) as being unpatentable over *Kanski* in view of *Maslov* as applied to claim 1 and further in view of *Kotoulas* (20020117579).